



Europäisches Patentamt
European Patent Office
Office européen des brevets



(11) Publication number:

0 489 712 A2

(12)

EUROPEAN PATENT APPLICATION

(21) Application number: **92101745.5**

(51) Int. Cl.⁵: **F42B 10/66**

(22) Date of filing: **10.02.89**

This application was filed on 03 - 02 - 1992 as a divisional application to the application mentioned under INID code 60.

(30) Priority: **11.02.88 GB 8803164**

(43) Date of publication of application:
10.06.92 Bulletin 92/24

(60) Publication number of the earlier application in accordance with Art.76 EPC: **0 329 342**

(84) Designated Contracting States:
DE FR GB IT

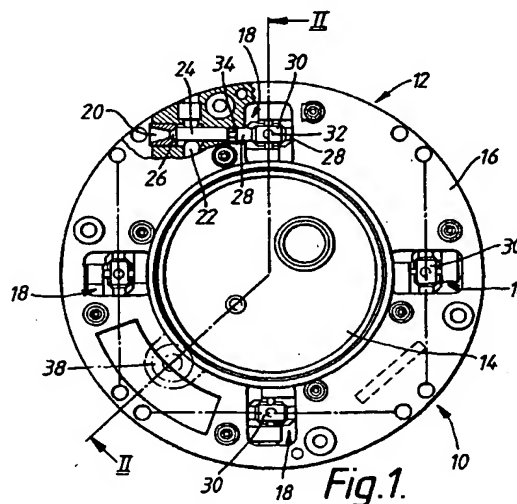
(71) Applicant: **BRITISH AEROSPACE PUBLIC LIMITED COMPANY**
11 Strand
London WC2N 5JT(GB)

(72) Inventor: **Machell, Anthony, c/o British Aerospace Plc.**
Six Hills Way
Stevenage SG1 2DA, Hertfordshire(GB)
Inventor: **Foulsham, Peter G., c/o British Aerospace Plc.**
Six Hills Way
Stevenage SG1 2DA, Hertfordshire(GB)

(74) Representative: **Newell, William Joseph et al**
Wynne-Jones, Laine & James 22 Rodney Road
Cheltenham GL50 1JJ(GB)

(54) **Missile steering arrangement using thrust control.**

(57) A reaction control system for a projectile includes two oppositely directed nozzle assemblies (20) each including a control element (24). The control elements are operated inversely by a control member (30). Each control element (24) includes lost motion means (34) to accommodate movement between a sealing end (26) and a distal end (28) thereof.



EP 0 489 712 A2

This invention relates to a reaction control system for a projectile, for example a missile or a mortar round and to projectiles including such systems.

Directional control of missiles in flight may be achieved either aerodynamically using movable control surfaces, e.g. fins, or by using a reaction control system in which control thrusts are generated by emitting a reaction gas transversely of the missile. In the known examples of the latter method of control it is usual to employ single shot squibs or pulsed units. This is believed to limit the degree and accuracy of control and requires the missile autopilot to be specifically designed to deal with this method of control. In these systems the magnitude of the thrust remains generally the same; the only control available is control of the length of the impulse. Studies conducted by the applicants show that there is a need for a reaction control system which produces a thrust whose magnitude may be varied continuously. Furthermore, there is a need for such a system in which the relationship between the movement of the member controlling the reaction control system and the thrust thereby is generally compatible with that of a fin assembly so that the reaction system may be controlled by an existing autopilot without major modification.

EP-A-244971 discloses a reaction control system in which lost motion is provided between a lever and a pinion in the drive train between an electromechanical actuator and a valve control element, for the express purpose of avoiding the need to match the strokes of the actuator and the valve.

According to one aspect of the invention, there is provided a reaction control system for a projectile comprising a nozzle assembly including a control element movable to control the flow therethrough, a control member for moving said control element and lost motion means in the control path between said control member and said control element characterised in that said control element comprises a sealing end and a distal end movable relative to each other, and said lost motion means comprises bias means arranged between said sealing end and said distal end, whereby said distal end may move relative to said sealing end.

The invention also extends to a projectile including the reaction control system as defined above.

In a further aspect of this invention, there is provided a projectile including a reaction control system as defined above capable of effecting control in each of the control senses of the projectile, aerodynamic control surface means movable for effecting control in each of the control senses of the projectile, and autopilot means for effecting control simultaneously via said reaction control sys-

tem and said aerodynamic control surface means.

A preferred embodiment of the system is particularly useful where a projectile requires the control function exerted by the aerodynamic fins either to be augmented by the reaction control system, e.g., to execute a terminal manoeuvre, or to be replaced thereby when the projectile is travelling too slowly for the aerodynamic fins to be effective, e.g. at launch.

The invention will now be described by way of non-limiting example, reference being made to the accompanying drawings, in which:-

Figure 1 is a schematic rear view of an actuator assembly of an example of reaction control system according to this invention;

Figure 2 is a section view of the actuator assembly of Figure 1 taken on lines II-II;

Figure 3 is a diagrammatic view of the arrangement of one pair of nozzles in the arrangement of Figures 1 and 2;

Figure 4 is a graph illustrating the relationship between piston deflection, thrust and fin deflection for an example of a missile incorporating a reaction control system and a fin assembly;

Figure 5 is a diagrammatic view of an example of a missile incorporating a reaction control system and a fin assembly;

Figure 6 is a block diagram of the control system of the missile of Figure 5;

Figures 7 and 8 show an example of a control thruster assembly in the open and closed positions respectively; and

Figure 9 shows schematically an example of a control thruster assembly configured in a servo loop.

Referring to the Figures, the illustrated embodiment of reaction control system is intended to be operated in conjunction with a fin assembly by an autopilot. The reaction control system may however also be used in missiles which do not incorporate a fin assembly.

Figures 1 and 2 illustrate an actuator assembly 10 for a reaction control system. The assembly 10 comprises a casing 12 which houses a gas generator 14 and which includes an outer flange 16. The outer flange carries four thruster assemblies 18 equispaced around the periphery of the missile. Each thruster assembly 18 comprises a pair of aligned nozzles 20 connected by conduits 22 to the gas generator 14 and each arranged to exhaust in opposite directions. The flow of gas from gas generator 14 through each nozzle 20 is controlled by means of a piston 24 slidably mounted with respect to the nozzle so as to be movable to vary the effective thrust area of the nozzle and thus the thrust generated. Each piston is continuously movable with respect to the throat of the associated nozzle - i.e. it can assume any position between

fully open and fully closed. Each piston is provided with a conical sealing end 26 and a distal end 28. The respective distal ends of the pistons 24 of a pair of nozzles are acted upon by a cam member 30 rotatably secured to the flange 16 and attached to an operating lever 32. Rotation of the cam member causes inverse operation of the pistons, i.e. one piston moves to increase the flow through its associated nozzle as the other piston moves to decrease the flow through its associated nozzle. A spring arrangement 34 is provided between the sealing end 26 and the distal end 28 and is arranged so that in normal operation there is no relative movement between the sealing end and the distal end, but if the sealing end 26 should seize and be prevented from movement, the distal end may move relative to the sealing end 28 against the bias of the spring 34 so that the cam member 30 is not prevented from rotating.

The four thruster assemblies 18 are equispaced around the periphery of the missile so that two assemblies (the upper and lower assemblies as viewed in Figure 1) lie in spaced planes parallel to the yaw plane and two assemblies (the left hand and right hand assemblies as viewed in Figure 1) lie in spaced planes parallel to the pitch plane. In use, as illustrated in Figure 5, the actuator assembly illustrated in Figures 1 and is mounted forwardly of the centre of gravity of the missile. Hence, when used in unison, the upper and lower assemblies effect control in the yaw sense and the left and right hand assemblies effect control in the pitch sense. If the upper and lower assemblies are not operated to generate the same magnitude of thrust in parallel directions, then a component of roll torque is generated. If the thrust generated by the upper and lower assemblies is equal and opposite then a simple roll torque will be generated. Similar considerations apply to operation of the left and right hand assemblies.

The gas generator may be of any suitable form; in the illustrated embodiment, it takes the form of a hot gas generator which is ignited by means of an igniter 36. A burster disc assembly 38 is provided for safety purposes.

Figure 3 illustrates schematically a single assembly showing the operating lever 32 and the cam 30. It will be understood that the direction and magnitude of the thrust vector generated by the assembly is dependent on the position of the operating lever 32.

Figure 4 illustrates the variation of the thrust developed with the deflection of the piston; it will be seen that the thrust varies proportionally with movement of the piston. It should be noted that this is for the purposes of illustration only and that other characteristics will result for different designs.

Figure 5 illustrates a missile with a reaction

control assembly 10 located forwardly of the centre of gravity 40 of the missile and an aft fin assembly 42 comprising four movable fins 44 arranged at the aft of a missile. The fins are oriented around the missile body so that operation of one set of diametrically opposed fins in unison effects control in the yaw sense whilst operation of the other set of diametrically opposed fins in unison effects control in the pitch sense. Differential operation of either set of fins effects control in the roll sense.

Figure 6 illustrates a navigation system for the missile of Figure 5. An autopilot 46 calculates the control movements required for the desired course corrections and controls a servomotor assembly 48 which controls movement of the fins 44 and also movement of the operating levers 32 of the associated thruster assembly. It will be understood that each of the thruster assemblies is operable to impart a control moment which is similar to that imparted by rotation of a fin member when the missile is in normal flight. Thus the control function exerted by thruster assembly 18' is analogous to that exerted by movable fin 44' etc. The thrust developed on deflection of the fin 44 is illustrated in Figure 4. It will be seen that, as with movement of the pistons of the thruster assemblies, that thrust/movement relationship is essentially linear for this example and that the thrust/movement characteristics for the thruster assembly are similar to those developed by angular movement of the fin.

Thus, by linking the operating levers 32 to a normal fin servo system, the reaction control system may provide a thrust which is proportional to the fin deflections, so enhancing the control effectiveness of the fins. This is particularly useful when the missile is travelling too slowly for the fins to be effective e.g. at launch, or where an extra amount of control is desired e.g. for a terminal manoeuvre. Because of the similarity between the movement/thrust characteristics of the fins and reaction control system this method of control augmentation may be added to a missile with little or no change to the autopilot.

In the arrangements of Figures 1 to 6, since the magnitude of the thrust developed can be adjusted across a large, continuous, range of values, the missile "weave" associated with squibs or pulsed thrusters can be avoided.

Where the reaction control system is employed in conjunction with a conventional movable fin assembly, the spring 34 override mechanism in the reaction control system serves an important purpose because it prevents total failure in the event of a failure of the reaction control system, as it enables the movable fin assembly to continue operating.

The spring override mechanism is of particular

benefit where two nozzles are operated in back to back fashion by a single actuator as in the arrangements 18 of Figures 1 to 6, because in these types of arrangement the mechanism may compensate for slight dimensional inaccuracies of the piston members, the cam mechanism and/or the housing defining the bores in which the pistons slide which might otherwise jam or damage the drive motor when moving towards an end position. The mechanism means that the designer can ensure that the pistons may be moved into engagement with the nozzle throat, thus closing it, without jamming the servo control system or preventing movement of the corresponding fin member.

The reaction gas may be a cold or hot gas and the storage reservoir or gas generator may be integrated with the nozzle units to provide a compact control package. The reaction gas may be bled off the main missile rocket motor; indeed a plurality of thruster assemblies may be inclined rearwards to provide not only lateral control but also the main source of rocket propulsion.

Referring now to the examples illustrated in Figures 7 to 9, these are intended to provide a control thruster assembly in which the force or torque required to move the control element is greatly reduced, so that the assembly is suitable for use in flight vehicles such as guided missiles.

Figures 7 and 8 illustrate an embodiment of control thruster assembly in the fully open and fully closed positions respectively. The assembly comprises a housing 50 defining an outlet throat 52 and a fluid supply manifold 54, and a plunger 56 which is slidably located in a pair of spaced bores 58, 60 each of which includes a gas seal 62. The bore 60 at the rear of the plunger 56 is closed to define a variable volume chamber 63, together with the rear end of the plunger 56. A pressure balance bore 64 interconnects the two axial end faces of the plunger 56 to tend to equalise the pressure forces acting on the plunger 56.

A pressure transducer 66 may be provided to sense the pressure in the variable volume chamber 63, the sensed pressure being a measure of the pressure of the fluid in the outlet throat 52 and also of the displacement of the plunger 56. Intermediate the bores 58 and 60, the housing 50 is relieved and receives for limited axial movement an actuating lever 68 secured to the plunger 56. The actuating lever 68 includes a rack portion 70 which engages the pinion 72 of a drive motor (not shown) for driving the plunger between the positions shown in Figures 7 and 8. A plunger position sensor linkage 74 may be connected to the actuating lever 68 for determining the position of the plunger.

In this arrangement, the thrust can be switched "on" or "off" or modulated by movement of the plunger 56. For a proportional thrust system the

drive may be in the form of a servo motor/gear box, whilst for a bang-bang system a stepping motor drive connected directly to the plunger may be used.

An important part of the assembly of Figures 7 and 8 is the pressure balance bore 64, which allows the pressure at the outlet throat to be sampled.

The magnitude of the sampled pressure will be a function of the plunger displacement. With this system only a modestly powered servo actuator drive will be required for the plunger as the resultant pressure force acting on the plunger will be low. In the "off" position, ambient pressure will act on both ends of the plunger.

Plunger position data may be determined from a position sensor connected to the position sensor linkage 74 or it may be derived from a pressure transducer 66 which samples the pressure at the outlet throat via the pressure balance bore 64.

For high gas temperature operation, refractory or ceramic materials may be used in the assembly. The gas supply to the thruster assembly may be taken from the main propulsion rocket motor system or from a dedicated hot or cool propellant gas supply. The number of thrusters will be a function of the particular control system required.

It will readily be seen that this arrangement of Figures 7 and 8 may be incorporated into the proportional control systems illustrated in Figures 1 to 6. Alternatively, it may be incorporated in a bang-bang system.

Figure 9 is a schematic illustration of a thruster assembly of the type illustrated in Figures 7 and 8 configured in a servo-control loop. Many of the component parts are similar and will not be described in detail again. In this arrangement the plunger 56 has an integral rack portion 70 which meshes with a pinion 72 connected via a gear train to an electric motor 80. A potentiometer 81 associated with the electric motor 80 provides a position feedback signal which is supplied to one input of a differential amplifier 82 which controls the motor 80.

The position servo can control a single unit as shown with a pressure balance bore 64 to reduce actuation forces, or two units arranged back-to-back as in the arrangements of Figures 1 to 6, where the pressure balance bore would not be necessary. Similarly a servo control loop of the type illustrated in Figure 9 could be used to drive the cam mechanism of the arrangements of Figures 1 to 6.

In the examples of Figures 7 to 9, the use of torque motors or gear motors to actuate the pistons means that it may be possible to transmit larger forces which might damage the cam-type arrangements of Figures 1 to 6.

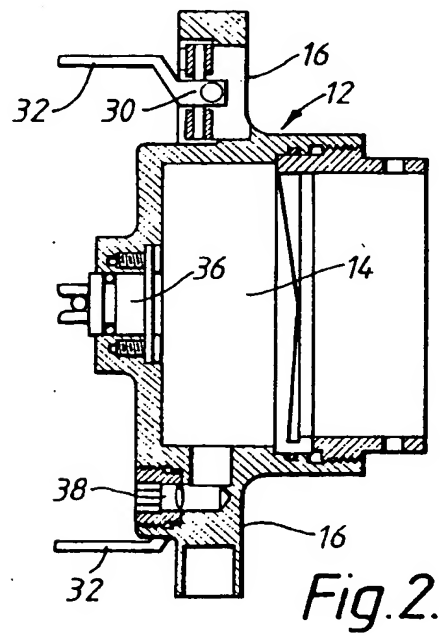
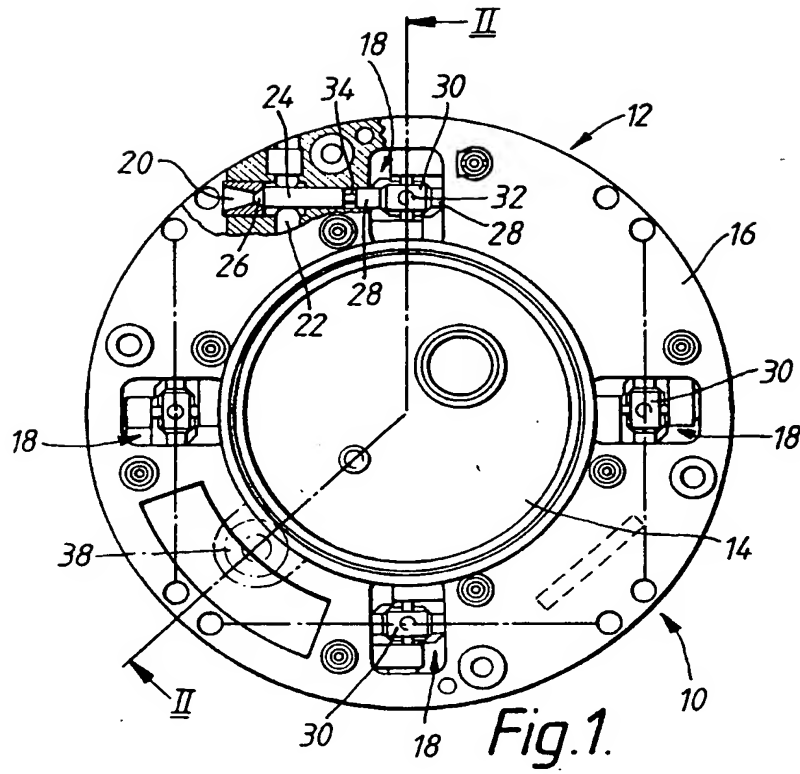
It will be understood that the reaction control systems described herein may be coupled to operate in tandem with the fin control system or they may operate independently of any fin control system.

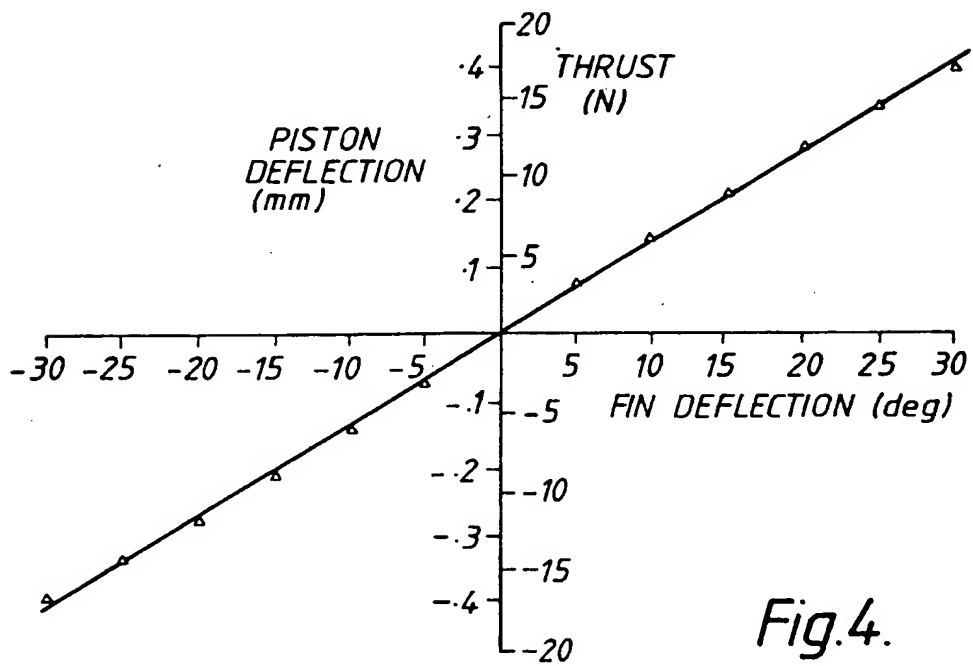
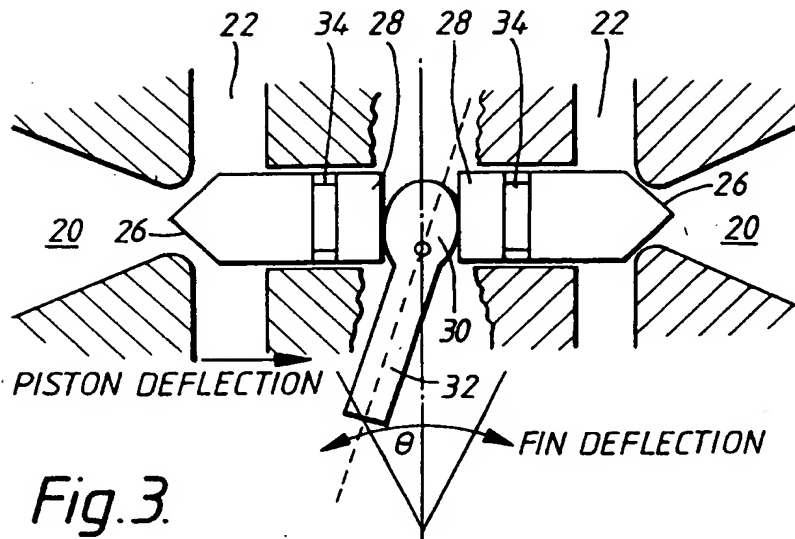
Claims

1. A reaction control system for a projectile comprising a nozzle assembly (20) including a control element (24) movable to control the flow therethrough, a control member (30) for moving said control element (24) and lost motion means (34) in the control path between said control member (30) and said control element (24) characterised in that said control element (24) comprises a sealing end (26) and a distal end (28) movable relative to each other and said lost motion means comprises bias means (34) arranged between said sealing end (26) and said distal end (28), whereby said distal end (28) may move relative to said sealing end (26). 10
2. A reaction control system according to Claim 1, wherein said bias means comprises a spring (34) located between said sealing end (26) and distal end (28). 15
3. A reaction control system according to Claim 1 or Claim 2, including a further oppositely directed nozzle assembly (20) including a further moveable control element (24) having a sealing end (26) and a distal end (28) and lost motion means (34) located between said sealing end (26) and said distal end (28), said further moveable control element (24) being acted upon by said control member (30). 20
4. A reaction control system according to any preceding claim, wherein the or each nozzle assembly (20) has an aerodynamic control surface means (44) operatively associated therewith for generating a control thrust or movement in generally the same sense as the control thrust generated in use by said nozzle assembly (20). 25
5. A reaction control system according to any preceding claim, wherein the or each control element (24) comprises a piston means (56) having a sealing face (26) which in use is exposed to the outlet pressure of said nozzle assembly (20). 30
6. A reaction control system according to Claim 5, wherein said piston means (56) includes passage means (64) for communicating the 35

pressure at said sealing face (26) to a distal face (28) of said piston means (56).

7. A reaction control system for a projectile comprising two oppositely directed nozzle assemblies (20) each including a control element (24) moveable to control the flow therethrough, a common control member (30) for moving said control elements (24), and lost motion means (34) for coupling said control member (30) and said control elements (24) characterised in that the lost motion means (34) comprises a spring (34) located between a sealing end (26) and a distal end (28) of each control element (24) so that each distal end (28) can move relative to each sealing end (26) against the bias of the spring (34). 40





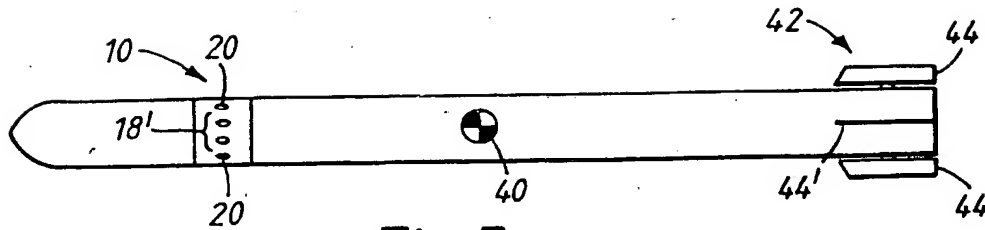


Fig. 5.

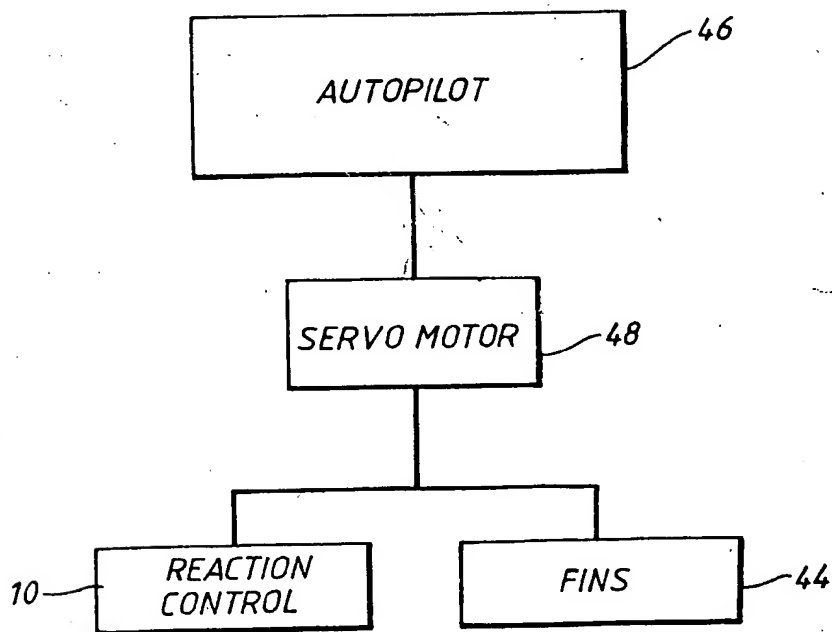
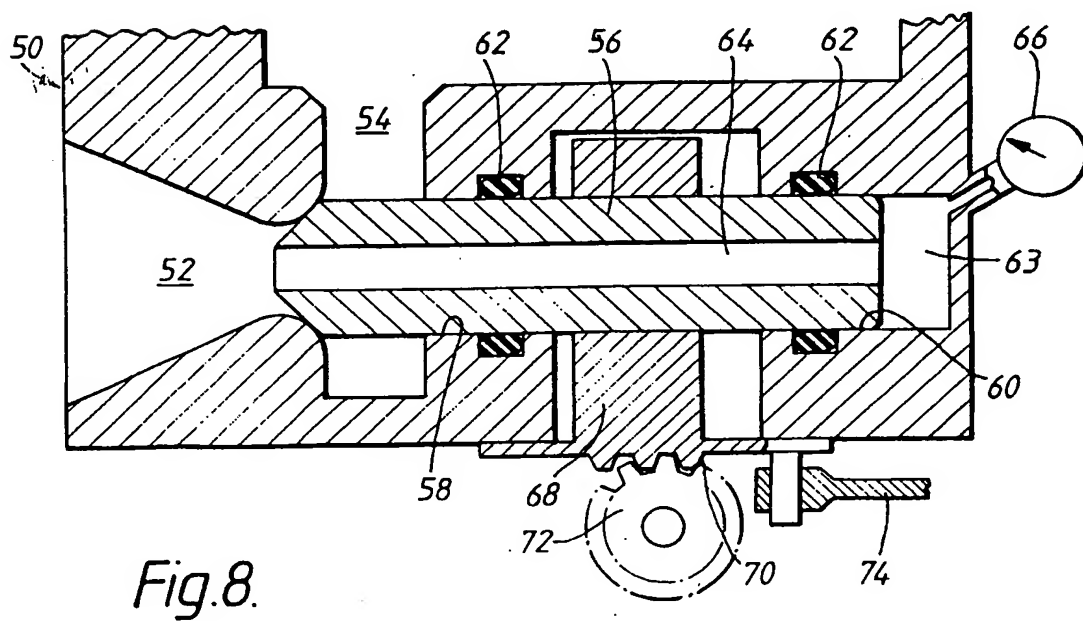
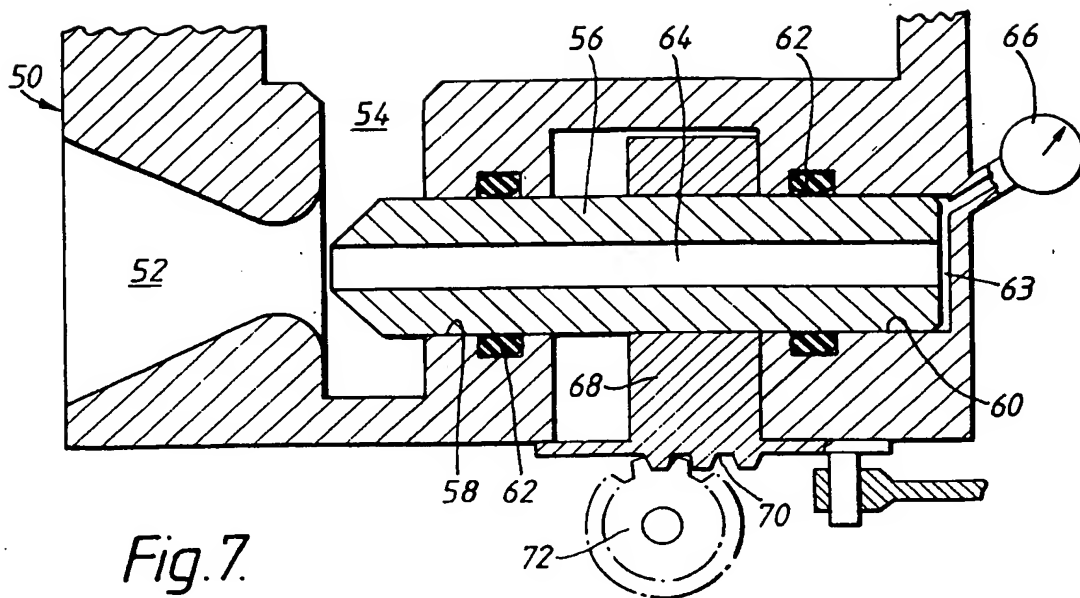


Fig. 6.



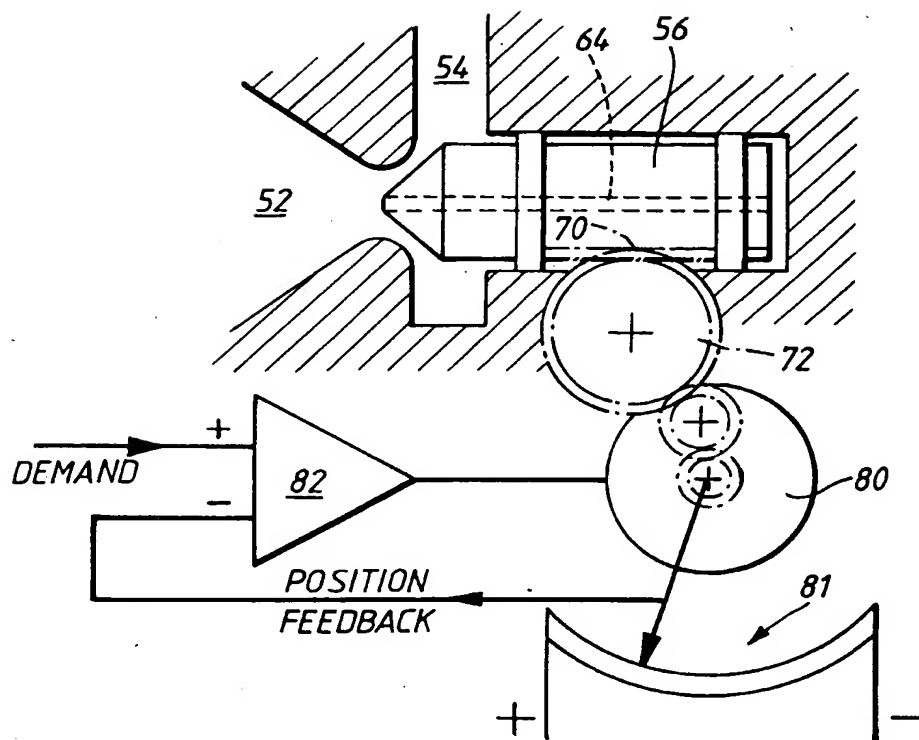


Fig.9.